* Effective exhaust velocity
  + C = v2 + (p2 – p3)\*A2/m
  + Thrust, F = mdot\*v2 + (p2– p3)\*A2
    - F = mdot\*c
  + Total Impulse
    - It = integral(0🡪t) F\*dt
    - Q-class: It = [18400, 36200] lbf
    - Specific impulse: Isp = It/mp/g0
* Mass Ratio
  + MR = mf/m0
  + Mass Prop Fraction: Xi = mp/m0
* Thrust to weight ratio: F/W0
  + Max at motor burnout
* Intrepid
  + It = 32181.171 lbf\*s
  + m0 = 348 lb
  + mp = 148 lb
  + Tt= 7.5s
  + mdot = 19.73 lb/s
  + F = 4290.8 lbf
  + Isp = 217.439 s
  + c = 6996.77 ft/s
* Rocket Equation
  + m\*dv/dt = -c\*dm/dt – mg
  + m\*dv/dt = -c\*mdot – mg
  + dv = -c\*dm/dt – g\*dt
  + v – v0 = -c\*ln(m(t)/m0) – g(t)